



INSTRUCTIONS FOR CONTINUED AIRWORTHINESS

For:
**CARGO/PASSENGER DOOR DOUBLER, TAILCONE, HORIZONTAL/VERTICAL
STAB AND WING STRUT FITTING/STRUCTURE REPLACEMENT
FOR TEXTRON AVIATION INC. 208 AND 208B AIRCRAFT**

Aircraft Structures International Corporation
620 N. Rock Rd. Suite 230, #233
Derby, KS 67037
Phone: 580-242-5907 Fax: 580-242-5834
www.asic.aero

Document No.: 16534000-600


Revision (B)

Date: 13MAR19

Applicable to:

Textron Aviation Inc. 208 and 208B

Prepared By: 
Martin McCaslin

Reviewed By: 
Daniel Garrett

The information in this Instruction for Continued Airworthiness is FAA accepted material and complies with 14 CFR 23.1529, Instructions for Continued Airworthiness. It supersedes or adds to that provided in the Maintenance Manual for the Textron Aviation Inc. 208 and 208B Aircraft, only where covered in the items contained herein. For limitations and procedures not contained in the Supplement, consult the Component Maintenance Manual, or other approved aircraft data.

620 N. Rock Rd. Suite 230, #233
Derby, KS 67037

13MAR19
Revision B

REVISION PAGE
Instructions for Continued Airworthiness

Updates to the ICA will be made by Aircraft Structures International Corp. Updates will be listed in the Log of Revisions and the List of Effective Pages below.

Log of Revisions				
REV. NO.	EFFECTED PAGE(S)	DESCRIPTION	DATE	APPROVED BY
Orig. Issue	All	Initial Release Rev (IR)	6OCT17	KMT
A	9	Revised details in Table 6	01NOV18	KMT
B	9-10	Added inspection tasks 53-10-00-258, 53-10-00-259, 55-30-00-251, and 55-10-00-252 and related details to Table 6	13MAR19	KMT

620 N. Rock Rd. Suite 230, #233
Derby, KS 67037

13MAR19
Revision B

List of Effective Pages

<u>Page</u>	<u>Date</u>	<u>Rev.</u>
Cover	13MAR19	B
1	13MAR19	B
2	13MAR19	B
3	13MAR19	B
4	13MAR19	B
5	13MAR19	B
6	13MAR19	B
7	13MAR19	B
8	13MAR19	B
9	13MAR19	B
10	13MAR19	B
11	13MAR19	B
12	13MAR19	B
13	13MAR19	B
14	13MAR19	B
15	13MAR19	B
16	13MAR19	B
17	13MAR19	B
18	13MAR19	B
19	13MAR19	B

Table of Contents

Log of Revisions	1
List of Effective Pages	2
Abbreviations & Definitions	4
1.0 INTRODUCTION	5
<i>Table 1 – Parts for Cargo/Passenger Door Doubler</i>	6
<i>Table 2 – Parts for Tailcone Fittings/Structure</i>	6
<i>Table 3 – Parts for Horizontal Stabilizer Fittings/Structure</i>	7
<i>Table 4 – Parts for Vertical Stabilizer Forward and Aft Spar</i>	8
<i>Table 5 – Parts for Strut Fittings</i>	8
2.0 INSPECTION REQUIREMENTS & OVERHAUL SCHEDULE	8
<i>Table 6 – Supplemental Inspection Requirements</i>	9
3.0 PART INSTALLATION TRACKING LOG	10
<i>Table 7 – Part Installation Tracking Log</i>	10
4.0 DIMENSION & ACCESS	13
5.0 LIFTING & SHORING	13
6.0 LEVELING & WEIGHING	13
7.0 TOWING & TAXIING	13
8.0 PARKING & MOORING	13
9.0 PLACARDS & MARKINGS	13
10.0 SERVICE INFORMATION	13
11.0 AIRWORTHINESS LIMITATIONS	14
APPENDIX A – INSTALLATION FIGURES	15
<i>Figure A-1 – Cargo/Passenger Door Doubler</i>	15
<i>Figure A-2 – Tailcone Fittings/Structure</i>	16
<i>Figure A-3 – Horizontal Stabilizer Fittings/Structure</i>	17
<i>Figure A-4 – Vertical Stabilizer Forward and Aft Spar</i>	18
<i>Figure A-5 – Wing Strut Fittings</i>	19

620 N. Rock Rd. Suite 230, #233
Derby, KS 67037

13MAR19
Revision B

Abbreviations & Definitions

Abbreviation	Definition
CMM	Textron Aviation Inc. Maintenance Manual
DVI	Detailed Visual Inspection
FAA	Federal Aviation Administration
HFEC	High Frequency Eddy Current
ICA	Instructions for Continued Airworthiness
MDL	Master Data List
NDI	Nondestructive Inspection
OEM	Original Equipment Manufacturer
SID	Supplemental Inspection Document
SRM	Structural Repair Manual
STC	Supplemental Type Certificate

620 N. Rock Rd. Suite 230, #233
Derby, KS 67037

13MAR19
Revision B

1.0 INTRODUCTION

This aircraft has been modified with a Supplemental Type Certificate that approves the replacement of certain Textron Aviation Inc. Aircraft Co. (Textron Aviation, Inc.) parts on Textron Aviation Inc. model 208/208B aircraft. Factory new parts have been used as replacements. Specifically, certain new Textron Aviation Inc. parts have been replaced on the aircraft that may be subject to a Supplemental Inspection Requirement for owner/operators complying with Chapter 5 of the Textron Aviation Inc. 208 Maintenance Manual. The parts replaced under this STC are subject to recurrent inspections when compliance with Chapter 5 is maintained; the inspection intervals for the replaced parts are reset to “zero time” intervals, while the inspection interval lengths and methods are unchanged.

The purpose of this Instructions for Continued Airworthiness (ICA) document is to provide the maintenance technician with the information necessary to ensure continued airworthiness of Textron Aviation Inc. 208/208B model aircraft modified in accordance with the installation information on Master Data List (MDL) 16534000-300 Rev IR or later FAA approved revision; specifically, these instructions define how to use the Textron Aviation Inc. Supplemental Structural Inspection Program after the aircraft has been modified.

Executing the modification defined by MDL 16534000-300 obligates the operator to include the maintenance information provided by this ICA into the operator's aircraft Maintenance Manual and operator's scheduled maintenance program. This document defines supplementary maintenance operations and frequencies recommended to ensure the aircraft's airworthiness.

The information contained herein addresses the requirements specified in 14 CFR 23.1529, ICA and supplements the basic aircraft maintenance manual only in those areas listed as pertains to this modification as installed per the MDL 16534000-300. For limitations and procedures not contained in this supplement, consult the basic aircraft maintenance manual.

Data

All information to support the continued airworthiness of this modification is contained in this ICA document.

For all aircraft, including those modified by the STC replacement fittings, the following Textron Aviation Inc. Maintenance Manual (CMM) 5-13-00 Supplemental Inspection Document statement applies:

The Supplemental Structural Inspection Program is valid for airplanes with less than 50,000 hours. Beyond this the continued airworthiness of the airplane can no longer be assured. Retirement of the airframe is recommended when 50,000 flight hours have been accumulated.

This STC does not in any way supersede or substantiate any deviation from the CMM 5-13-00 requirement. Therefore, the airworthiness of all aircraft modified by this STC will remain limited to 50,000 flight hours, as prescribed above.

Modification Procedure 16534000-001 Rev IR or later FAA approved revision, using STC replacement fittings and parts and conducted on Textron Aviation Inc. 208 and 208B aircraft, provides instructions for the replacement of specified fuselage fittings and parts for the cargo door doublers, empennage horizontal stabilizer, vertical stabilizer and struts with new Textron Aviation Inc. Original Equipment Manufacturer (OEM) parts.

The modification procedure resets the inspection requirements to “**Zero Time Since New**” for replaced fittings and parts on Textron Aviation Inc. 208 S/Ns 20800001 and on, and 208B S/Ns 208B0001 and on, resetting inspection requirements to the initial inspection interval regardless of aircraft total time in service. Subsequent inspections of the modified structure are set forth in document no. 16534000-600 ICA.

Design Change Control

All associated data and changes to the aircraft design are specified on MDL 16534000-300 Rev IR or later FAA approved revision. Revisions to this MDL are controlled by the STC holder listed on the front of FAA form 8110-2.

620 N. Rock Rd. Suite 230, #233
Derby, KS 67037

13MAR19
Revision B

The 16534000-300 MDL applies to all the following new OEM parts and assemblies on the Textron Aviation Inc. 208 and 208B aircraft.

Applicable Aircraft / Parts
Part Number/Serial Number Effectivity

Table 1 – Parts for Cargo/Passenger Door Doubler

PART NUMBER	SERIAL NUMBER	DESCRIPTION
2611088-1	20800001 AND ON	DOUBLER-LH OUTER STA 234.00
2611088-2	208B0001 AND ON	DOUBER-RH OUTER STA 259.00
2611088-3	20800001 AND ON	DOUBLER-LH OUTER STA 284.00
2611088-4	208B0001 AND ON	DOUBLER-RH OUTER STA 284.00

Table 2 – Parts for Tailcone Fittings/Structure

PART NUMBER	SERIAL NUMBER	DESCRIPTION
2612077-1	20800001 AND ON 208B0001 AND ON	FITTING-RH STABILIZER ATTACH
2612077-2	20800001 AND ON 208B0001 AND ON	FITTING-LH STABILIZER ATTACH
2612087-1	20800001 AND ON 208B0001 AND ON	FITTING-FWD LH STABILIZER ATTACH
2612087-2	20800001 AND ON 208B0001 AND ON	FITTING-FWD RH STABILIZER ATTACH
2612088-1	20800001 AND ON 208B0001 AND ON	FITTING-AFT LH STABILIZER ATTACH
2612088-2	20800001 AND ON 208B0001 AND ON	FITTING-AFT RH STABILIZER ATTACH
2612059-1	20800001 AND ON 208B0001 AND ON	BULKHEAD ASSY CANTED STA 475.88
2612030-34	20800001 AND ON 208B0001 AND ON	BULKHEAD-UPR
2612030-17	20800001 AND ON 208B0001 AND ON	DOUBLER-UPR
2612030-18	20800001 AND ON 208B0001 AND ON	DOUBLER-LWR
2612030-32	20800001 AND ON 208B0001 AND ON	DOUBLER-UPR

620 N. Rock Rd. Suite 230, #233
Derby, KS 67037

13MAR19
Revision B

Table 3 – Parts for Horizontal Stabilizer Fittings/Structure

PART NUMBER	SERIAL NUMBER	DESCRIPTION
2632018-1	20800001 AND ON 208B0001 AND ON	ATTACH FITTING
2632023-2	20800001 AND ON 208B0001 AND ON	CAP FWD UPPER
2632023-3	20800001 AND ON 208B0001 AND ON	CAP AFT UPPER
2632023-4	20800001 AND ON 208B0001 AND ON	CAP FWD LOWER
2632023-5	20800001 AND ON 208B0001 AND ON	CAP AFT LOWER
2632023-6	20800001 THRU 20800314 208B0001 TO 208B0777	WEB INBD
2632023-10	20800315 AND ON, 208B0778 AND ON	WEB INBD
2632023-7	20800001 AND ON 208B0001 AND ON	WEB OUTBD
2632023-8	20800001 AND ON 208B0001 AND ON	SPLICE PLATE
2632024-2	20800001 AND ON 208B0001 AND ON	WEB INBD
2632024-3	20800001 AND ON 208B0001 AND ON	WEB OUTBD
2632024-4	20800001 AND ON 208B0001 AND ON	CAP FWD UPPER, FWD LOWER
2632024-5	20800001 AND ON 208B0001 AND ON	CAP AFT UPPER, AFT LOWER
2632024-6	20800001 AND ON 208B0001 AND ON	SPLICE PLATE
2632027-2	20800001 THRU 20800330 208B0001 TO 208B0863	SUPPORT AFT
2632027-10	20800331 AND ON 208B0864 AND ON	SUPPORT AFT
2632027-3	20800001 THRU 20800330 208B0001 TO 208B0863	SUPPORT FWD
2632027-11	20800331 AND ON, 208B0864 AND ON	SUPPORT FWD

620 N. Rock Rd. Suite 230, #233
Derby, KS 67037

13MAR19
Revision B

Table 4 – Parts for Vertical Stabilizer Forward and Aft Spar

PART NUMBER	SERIAL NUMBER	DESCRIPTION
2631021-15	20800001 AND ON 208B0001 AND ON	WEB
2631021-14	20800001 AND ON 208B0001 AND ON	SPAR CAP-RH
2631021-13	20800001 AND ON 208B0001 AND ON	SPAR CAP-LH
2631022-2	20800001 AND ON 208B0001 AND ON	WEB
2631022-5	20800001 AND ON 208B0001 AND ON	SPAR CAP-LH
2631022-6	20800001 AND ON 208B0001 AND ON	SPAR CAP-RH

Table 5 – Parts for Strut Fittings

PART NUMBER	SERIAL NUMBER	DESCRIPTION
2621008-201	20800001 THRU 20800179 208B0001 THRU 208B0209	FITTING-WING STRUT UPPER
2621008-202	20800180 AND ON 208B0210 AND ON	FITTING-WING STRUT UPPER
2621009-1	20800001 THRU 20800179 208B0001 THRU 208B0209	FITTING-WING STRUT LOWER
2621009-2	20800180 AND ON 208B0210 AND ON	FITTING-WING STRUT LOWER

2.0 INSPECTION REQUIREMENTS & OVERHAUL SCHEDULE

Installation of this STC does not change the aircraft manufacturer's maintenance methods and/or initial/recurrent inspection intervals for the affected structure. Installation of the STC does, however, reset the inspection interval to "zero time" for the affected components.

The initial and recurring inspection requirements for the OEM fittings and structure, listed in Table 6 column (1), are documented in the CMM 5-14-00 Listing of Supplemental Inspections. Table 6 provides a summary of the applicable data from CMM 5-14-00, as well as inspection intervals detailed in the inspection documents listed in column (5). The inspection documents listed in column (5) are detailed in the CMM 5-10-00. The current inspections require a Nondestructive Inspection (NDI) using HFEC probes. The Supplemental Inspection Number listed in Table 6 column (3) correlates to the appropriate document number in CMM Part 6 'Eddy Current' of the Nondestructive Testing Manual, detailing the eddy current inspection procedures. Column (2) lists the inspection task number NDI documents.

Table 6 will be updated if/when an update in intervals or procedures is issued by Textron Aviation Inc. that pertains to the affected structure.

620 N. Rock Rd. Suite 230, #233
Derby, KS 67037

13MAR19
Revision B

Table 6 – Supplemental Inspection Requirements

(1) Aircraft Structure Affected	(2) MM Inspection Task	(3) Supplemental Inspection Number	(4) Inspection Compliance Title	(5) Inspection Document	(6) Initial Inspection (Recurring Inspection) ¹
2611088-1/-2/-3/-4 Doubler Outer and Supporting Structure	53-10-00-251	53-20-01	Cargo and Passenger Door Doublers - Special Detailed Inspection	05-15-15	7,500 Hrs (2,500 Hrs)
2612077-1/-2 Fitting-Stabilizer Attach and Supporting Structure	53-10-00-257	53-50-01	Fuselage to Horizontal Stabilizer Attach Fittings - Special Detailed Inspection	05-15-13	20,000 Hrs (5,000 Hrs)
2612059-1, 2612030-17/-18/-32/-34 Bulkheads and Doublers for Vertical Stabilizer Attach	53-10-00-258	53-50-02	Vertical Stabilizer Attach Points Special Detailed Inspection (Typical Inspection Compliance)	05-15-13	20,000 Hrs (5,000 Hrs)
2612059-1, 2612030-17/-18/-32/-34 Bulkheads and Doublers for Vertical Stabilizer Attach	53-10-00-259	53-50-02	Vertical Stabilizer Attach Points Special Detailed Inspection (Severe Inspection Compliance)	05-15-17	16,500 Hrs (5,000 Hrs)
2632018-1 Attach Fitting Stabilizer and Supporting Structure	55-10-00-250	55-10-01	Horizontal Stabilizer Forward and Aft Attach Points -Special Detailed Inspection	05-15-13	20,000 Hrs (5,000 Hrs)
2612087-1/-2 Fitting-Fwd Stabilizer Attach and Supporting Structure	55-10-00-250	55-10-01	Horizontal Stabilizer Forward and Aft Attach Points -Special Detailed Inspection	05-15-13	20,000 Hrs (5,000 Hrs)
2612088-1/-2 Fitting-Aft Stabilizer Attach and Supporting Structure	55-10-00-250	55-10-01	Horizontal Stabilizer Forward and Aft Attach Points -Special Detailed Inspection	05-15-13	20,000 Hrs (5,000 Hrs)
2631021-13/-14/-15, 2631022-2/-5/-6 Spar Cap/Web and Supporting Structure	55-30-00-250	55-30-01	Vertical Stabilizer Spars - Special Detailed Inspection (Typical Inspection Compliance)	05-15-13	20,000 Hrs (5,000 Hrs)
2631021-13/-14/-15, 2631022-2/-5/-6 Spar Cap/Web and Supporting Structure	55-30-00-251	55-30-01	Vertical Stabilizer Spars Special Detailed Inspection (Severe Inspection Compliance)	05-15-17	16,500 Hrs (5,000 Hrs)
2632023-2/-3/-4/-5 2632024-4/-5 Cap-Fwd and Aft Upper, and Lower	55-10-00-251	55-10-02	Horizontal Stabilizer Spars - Special Detailed Inspection (Typical Inspection Compliance)	05-15-13	20,000 Hrs (5,000 Hrs)
2632023-2/-3/-4/-5 2632024-4/-5 Cap-Fwd and Aft Upper, and Lower	55-10-00-252	55-10-02	Horizontal Stabilizer Spars Special Detailed Inspection (Severe Inspection Compliance)	05-15-18	17,500 Hrs (1,000 Hrs)
2621008-1/-2/-201/-202 Upper Wing Strut Fitting	57-10-01-250	57-60-01	Upper Wing Strut Fitting - Special Detailed Inspection (Typical Inspection Compliance)	05-15-MA	10,000 Hrs (5,000 Hrs)
2621008-1/-2/-201/-202 Upper Wing Strut Fitting	57-10-01-251	57-60-01	Upper Wing Strut Fitting - Special Detailed Inspection (Severe Inspection Compliance)	05-15-ME	5,000 Hrs (3,600 Hrs)
2621009-1/-2 Lower Wing Strut Fitting	57-10-01-250	57-60-01	Lower Wing Strut Fitting - Special Detailed Inspection (Typical Inspection Compliance)	05-15-MA	10,000 Hrs (5,000 Hrs)

620 N. Rock Rd. Suite 230, #233
Derby, KS 67037

13MAR19
Revision B

(1) Aircraft Structure Affected	(2) MM Inspection Task	(3) Supplemental Inspection Number	(4) Inspection Compliance Title	(5) Inspection Document	(6) Initial Inspection (Recurring Inspection) ¹
2621009-1/-2 Lower Wing Strut Fitting	57-10-01-251	57-60-01	Lower Wing Strut Fitting - Special Detailed Inspection (Severe Inspection Compliance)	05-15-ME	5,000 Hrs (3,600 Hrs)

Note:

1. Initial and recurring inspection intervals are listed for aircraft operating under typical flight profiles. For maintenance of all aircraft affected by this STC, review the CMM, 4-00-00 and 5-13-00, to determine operational classification as typical or severe. Use severe classification inspection schedule specified in the CMM when aircraft operations are determined to be severe.

All structural fasteners are identified in Modification Procedure 16534000-001 Rev IR or later FAA approved.

3.0 PART INSTALLATION TRACKING LOG

The following parts may be installed in the aircraft per this STC. The STC may be applied to the aircraft in whole or in part. As such, the *Date Installed* and *TTIS* columns provides a means to log which parts have been installed in the aircraft and when the installation took place. Figures appear in the appendix for reference. The following table *must* be incorporated into the aircraft logs, and shall be maintained if/when listed parts are replaced.

Table 7 – Part Installation Tracking Log

Aircraft Model	<input type="checkbox"/> 208 <input type="checkbox"/> 208B
Registration Number	
Serial Number	

Part Number	Part Description	Date Installed	Aircraft TTIS at time of installation	Figure Reference
2611088-1	DOUBLER 0.040 2024-T42			A-1
2611088-2	DOUBLER 0.040 2024-T42			A-1
2611088-3	DOUBLER 0.040 2024-T42			A-1
2611088-4	DOUBLER 0.040 2024-T42			A-1
2612030-32	DOUBLER UPPER			A-2
2612030-18	DOUBLER LOWER			A-2
2612030-17	DOUBLER UPPER			A-2
2612030-34	BULKHEAD UPPER			A-2
2612059-1	BULKHEAD ASSY CANTED			A-2
2612088-2	FITTING-STABILIZER ATTACH AFT RH			A-2
2612088-1	FITTING-STABILIZER ATTACH AFT LH			A-2

620 N. Rock Rd. Suite 230, #233
Derby, KS 67037

13MAR19
Revision B

Part Number	Part Description	Date Installed	Aircraft TTIS at time of installation	Figure Reference
2612087-2	FITTING-STABILIZER ATTACH FWD RH			A-2
2612087-1	FITTING-STABILIZER ATTACH FWD LH			A-2
2612077-2	FITTING-STABILIZER ATTACH LH			A-2
2612077-1	FITTING-STABILIZER ATTACH RH			A-2
2632018-1	ATTACH FITTING			A-3
2632023-2	CAP FWD UPPER			A-3
2632023-3	CAP AFT UPPER			A-3
2632023-4	CAP FWD LOWER			A-3
2632023-5	CAP AFT LOWER			A-3
2632023-6	WEB INBD			A-3
2632023-10	WEB INBD			A-3
2632023-7	WEB OUTBD			A-3
2632023-8	SPLICE PLATE			A-3
2632024-2	WEB INBD			A-3
2632024-3	WEB OUTBD			A-3
2632024-4	CAP FWD UPPER, FWD LOWER			A-3
2632024-5	CAP AFT UPPER, AFT LOWER			A-3
2632024-6	SPLICE PLATE			A-3
2632027-2	SUPPORT AFT			A-3
2632027-10	SUPPORT AFT			A-3
2632027-3	SUPPORT FWD			A-3
2632027-11	SUPPORT FWD			A-3
2631022-6	SPAR CAP-RH			A-4
2631022-5	SPAR CAP-LH			A-4
2631022-2	WEB			A-4
2631021-13	SPAR CAP-LH			A-4
2631021-14	SPAR CAP-RH			A-4
2631021-15	WEB			A-4

620 N. Rock Rd. Suite 230, #233
Derby, KS 67037

13MAR19
Revision B

Part Number	Part Description	Date Installed	Aircraft TTIS at time of installation	Figure Reference
2621008-201	FITTING-WING STRUT UPPER			A-5
2621008-202	FITTING-WING STRUT UPPER			A-5
2621009-1	FITTING-WING STRUT LOWER			A-5
2621009-2	FITTING-WING STRUT LOWER			A-5

620 N. Rock Rd. Suite 230, #233
Derby, KS 67037

13MAR19
Revision B

4.0 DIMENSION & ACCESS

This modification procedure does not change the dimensions of the aircraft.

5.0 LIFTING & SHORING

No change

6.0 LEVELING & WEIGHING

No change

7.0 TOWING & TAXIING

No change

8.0 PARKING & MOORING

No change

9.0 PLACARDS & MARKINGS

No change

10.0 SERVICE INFORMATION

No change

620 N. Rock Rd. Suite 230, #233
Derby, KS 67037

13MAR19
Revision B

11.0 AIRWORTHINESS LIMITATIONS

The information contained herein supplements the basic Maintenance Manuals only in those areas listed, when the aircraft is modified in accordance with MDL 16534000-300 Rev IR or later FAA approved revision. For limitations and procedures not contained in this supplement, consult the basic Maintenance Manuals.

The Airworthiness Limitations section is FAA approved and specifies maintenance required under 14 CFR Sec. 43.16 and 91.403 of the Federal Aviation Regulations unless an alternative program has been FAA approved.

Fatigue Life-Limited Parts

Life limits are imposed on the strut channels within the strut assembly when the strut fittings are replaced per 16534000-100 Modification Procedure, Rev IR, or later FAA approved. The life limits on the strut channels are defined by total time on the struts since new and are irrespective of when the strut fittings are replaced or inspected. Replacement intervals are as follows:

Part Number	Description	Maximum Service Life (Typical Operations)	Maximum Service Life (Severe Operations)
2621000-17	Strut Channel Assembly	38,892 hrs	17,177 hrs

Definitions of typical and severe operations remain as defined in the Textron Aviation Model 208 Maintenance Manual, Chapter 4.

Approved by: _____ Date: _____
 Manager, Federal Aviation Administration
 Wichita Aircraft Certification Office

Distribution:

Per the requirement of 14 CFR Part 23.1529, changes made to the ICA by the applicant will be distributed via US Post by means of paper copy. A list will be maintained with the mailing address of each person authorized to use the STC. Each person installing this STC must notify the STC holder listed on the face of the STC, and must provide their name and mailing address for transmission of future updates.

APPENDIX A – INSTALLATION FIGURES

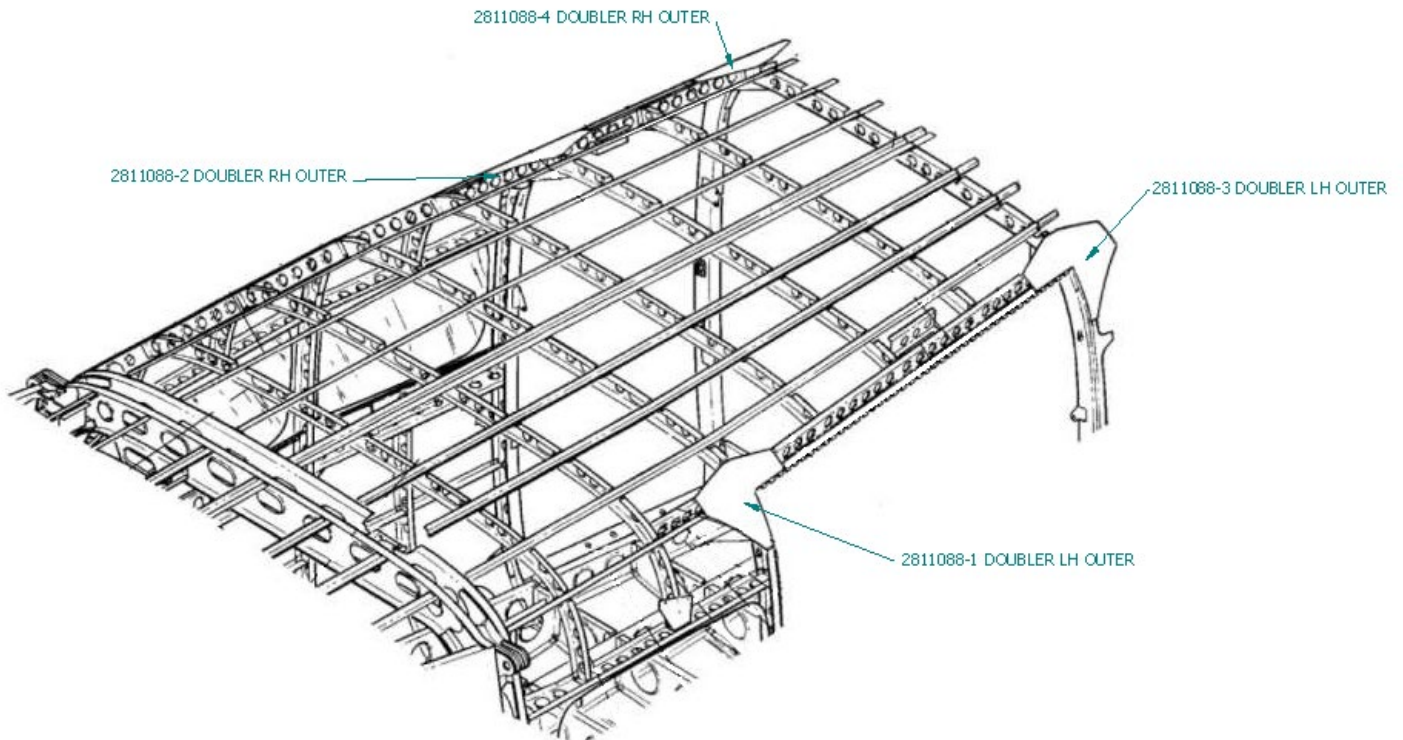


Figure A-1 – Cargo/Passenger Door Doubler

620 N. Rock Rd. Suite 230, #233
 Derby, KS 67037

13MAR19
 Revision B

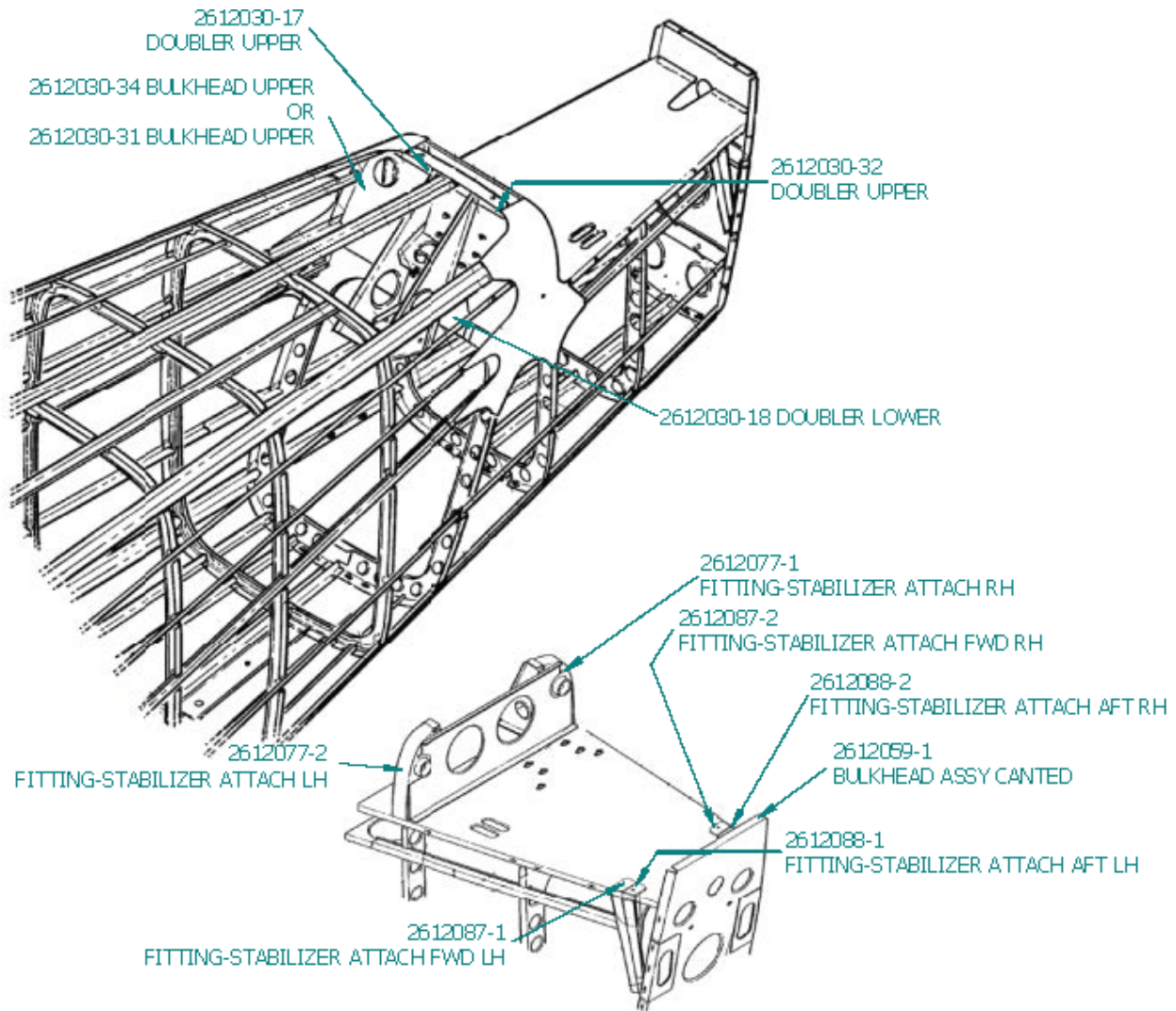


Figure A-2 – Tailcone Fittings/Structure

620 N. Rock Rd. Suite 230, #233
 Derby, KS 67037

13MAR19
 Revision B

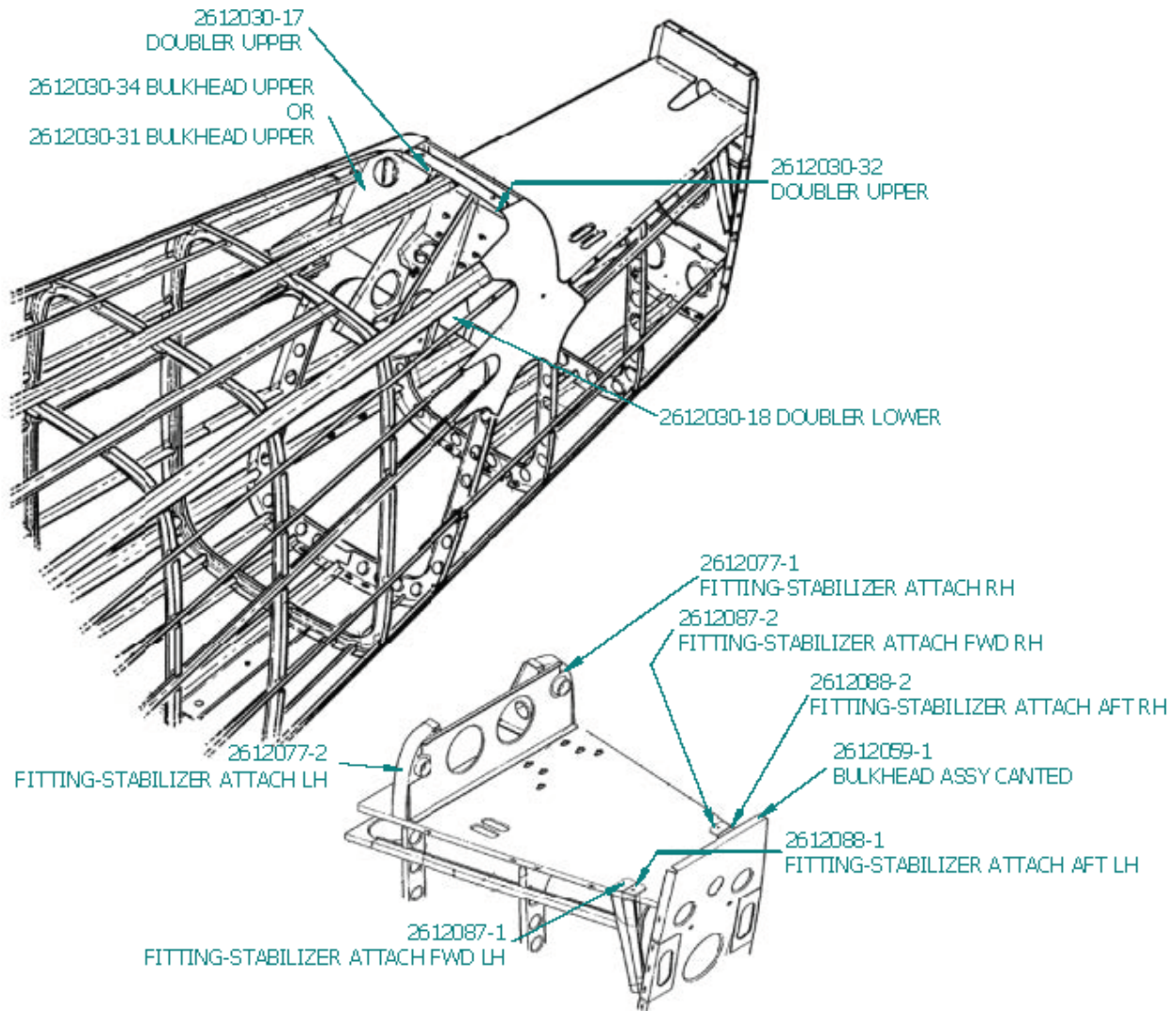


Figure A-3 – Horizontal Stabilizer Fittings/Structure

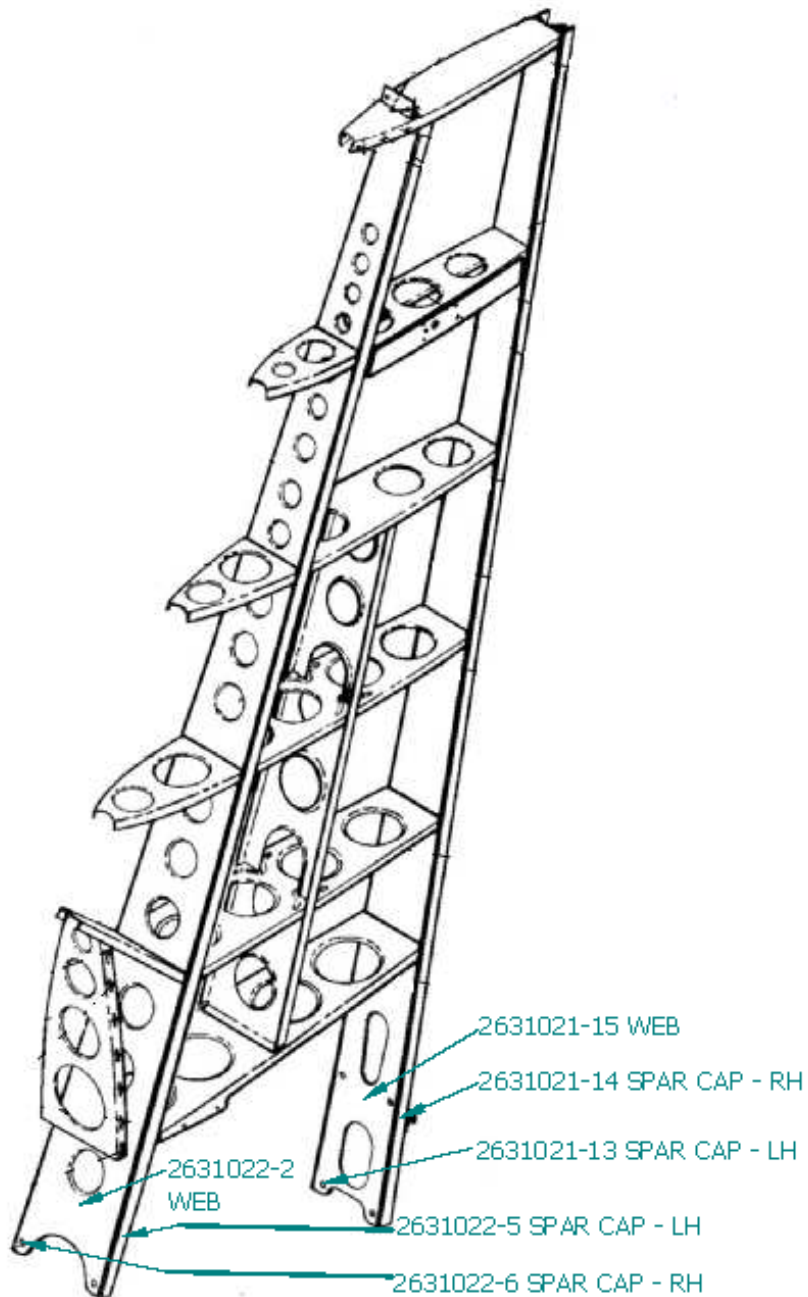


Figure A-4 – Vertical Stabilizer Forward and Aft Spar

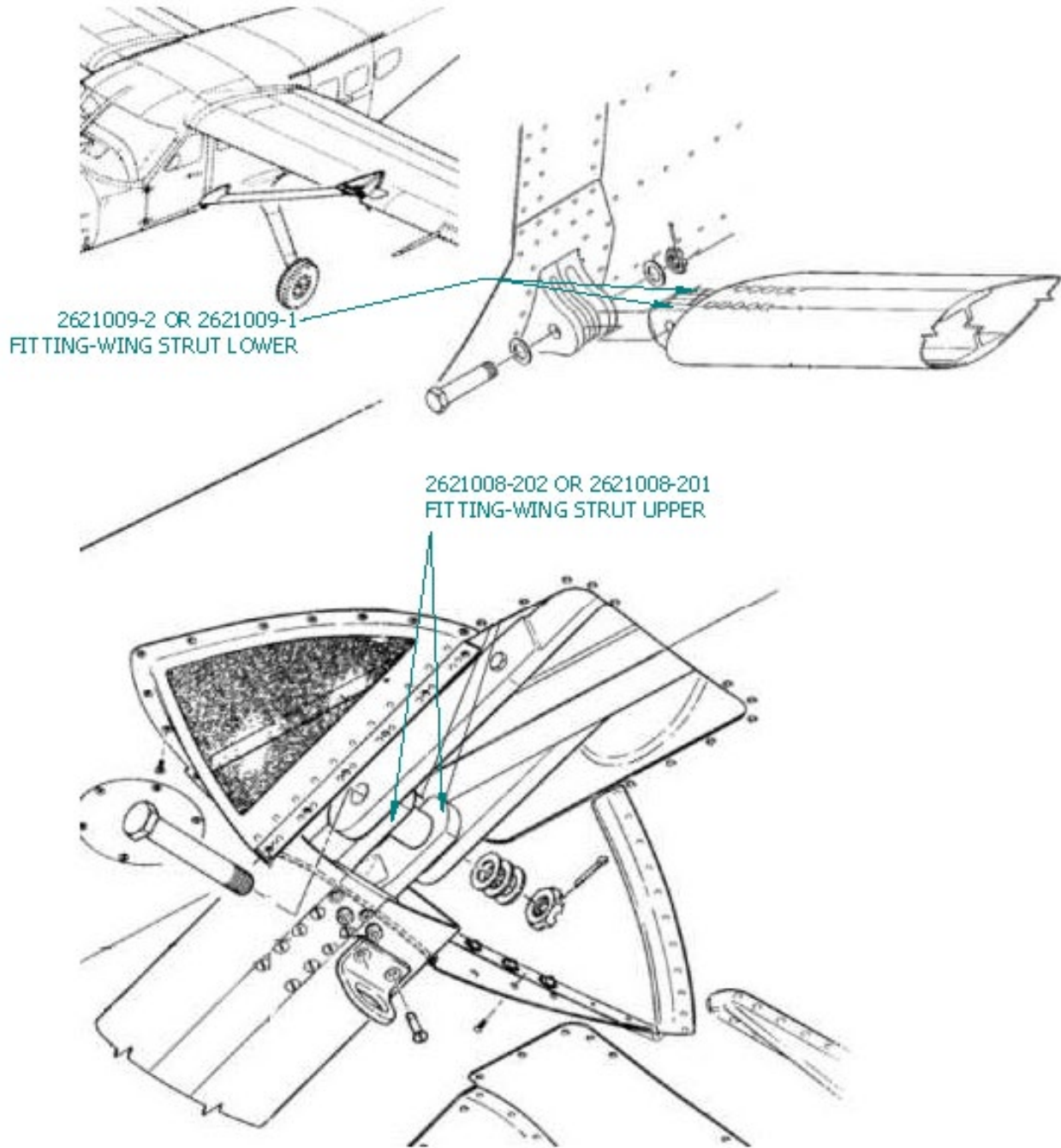


Figure A-5 – Wing Strut Fittings